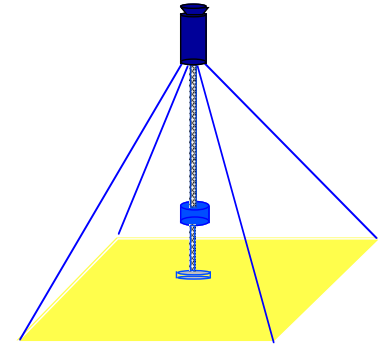




Conceptual Study of SSPS Demonstration Experiment

- Objectives of the study
- System configuration
- Flight operation
- Associated technologies



July 2003



Strategy and Policy

Strategy

First Step

Concept Development
for Practical SSPS



Second Step

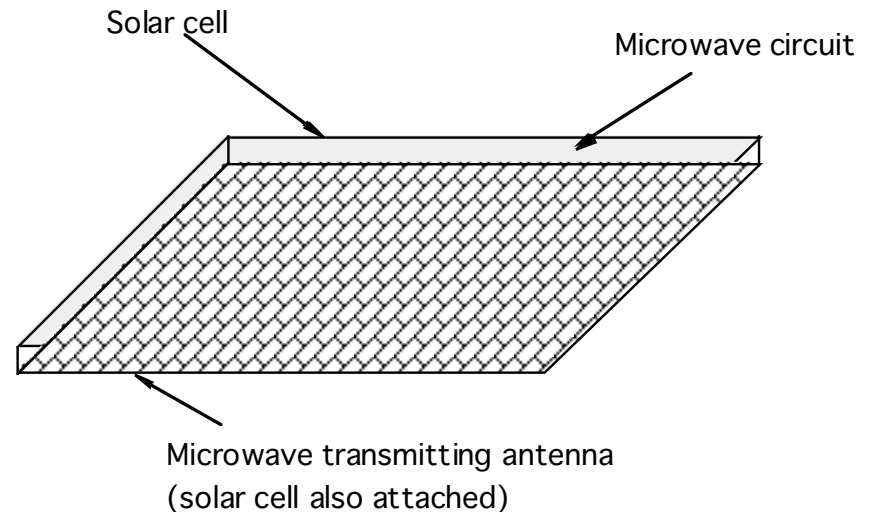
Design of
Demonstration SSPS

Policy 0

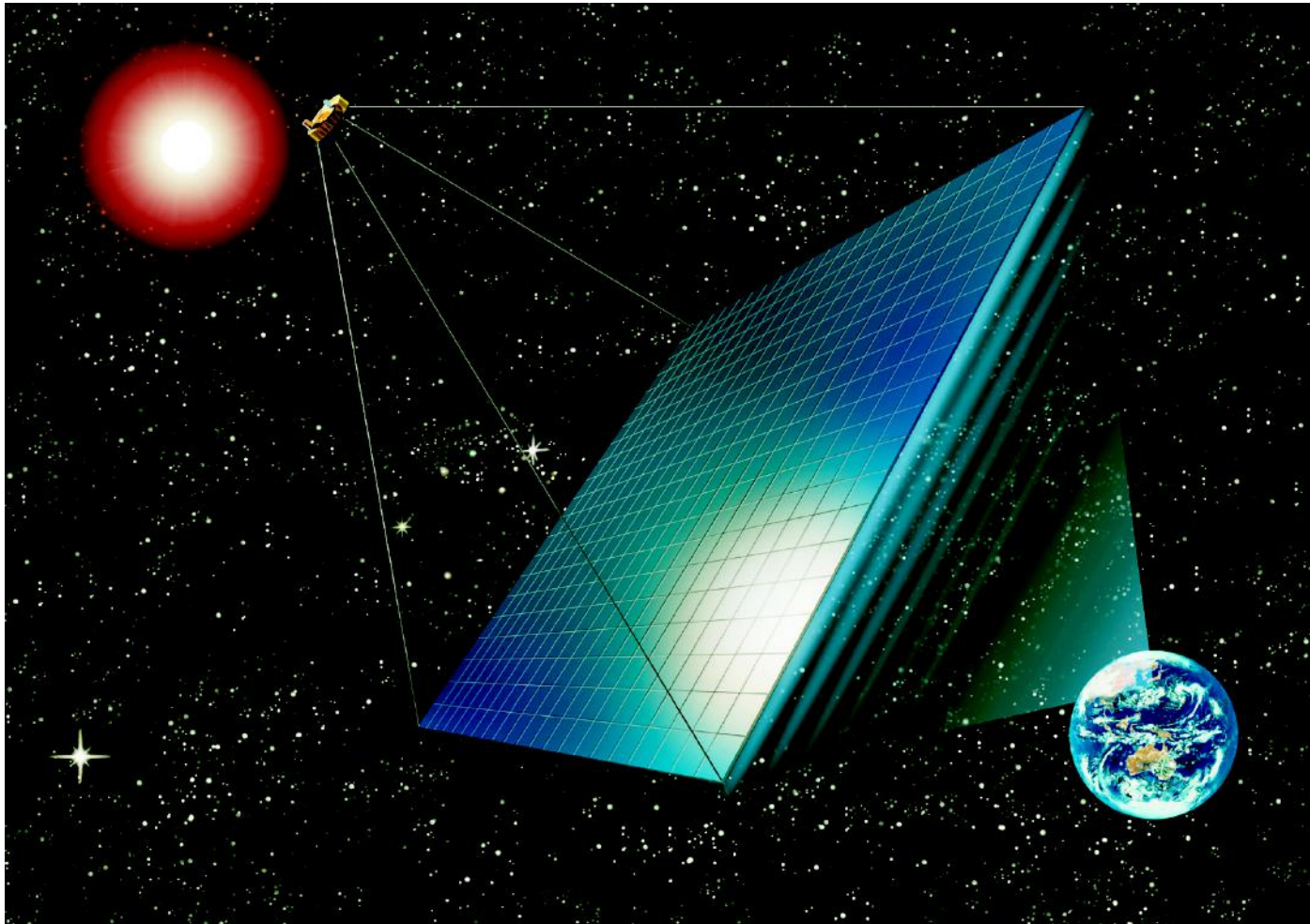
- A large number of electrical parts (solar cells, microwave elements) required: 0
 - ⇒ COTS (commercial off-the-shelf), high-technology, mass production, low-cost 0
- Structure and mechanism be simple and robust: 0
 - ⇒ no active attitude control and no movable mechanism in operation 0
- Distributed power system : 0
 - ⇒ power generation/transmission module (sandwich power module), no power bus 0
- Concentration of information: 0
 - ⇒ wireless LAN 0

Concept of Power Generation and Transmission Module (Sandwich Panel) 0

- Module including power generation and transmission 0
- Wireless interface between modules 0
- easy attachment and detachment as a module 0
- robust as a power system 0
- easy fabrication, test, integration, and maintenance



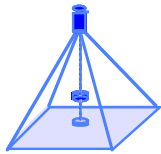
Practical or Commercial SSPS 0



Evolution of Demonstration Experiment to Practical SSPS 0

Commercial Model

Demonstration Experiment



Development



10 km



Bus

Power generator
(upper plane)

2.4 km

Power transmitter
(lower plane)

2.6 km

Microwave

Basic concept for
development

Power generation/
transmission module

Power generation/
transmission module

Wireless interface



Concept of Demonstration Experiment 0

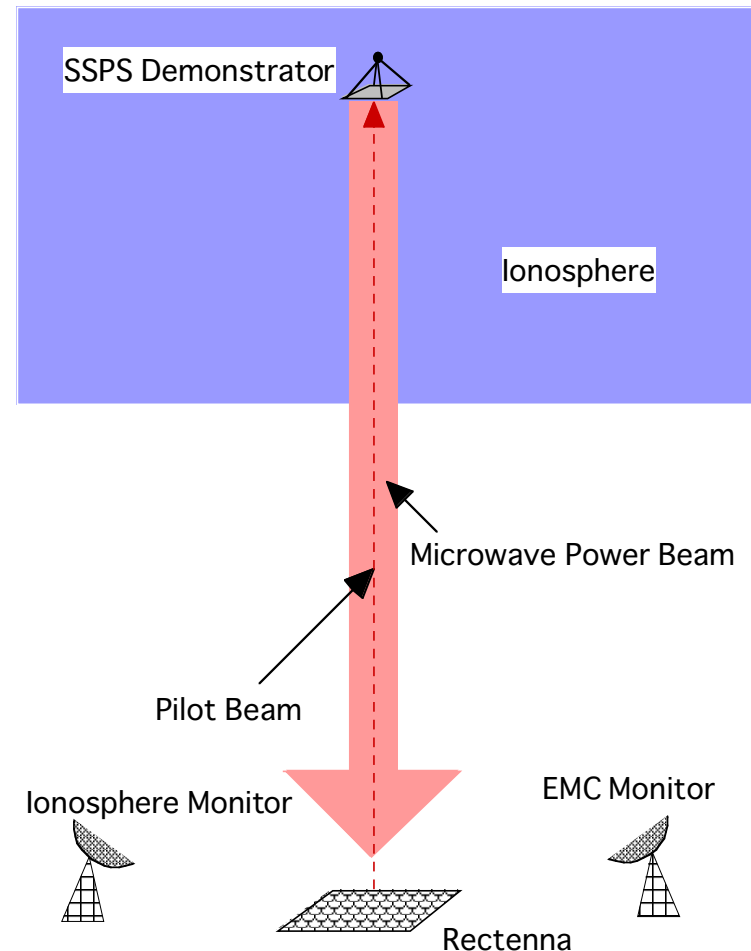
Most important subject at this stage towards the practical SSPS is a verification of power transmission from space to ground. 0

(1) Demonstration of microwave beam control, pointing precisely to a rectenna on ground from a large antenna dynamically-moving in orbit. 0

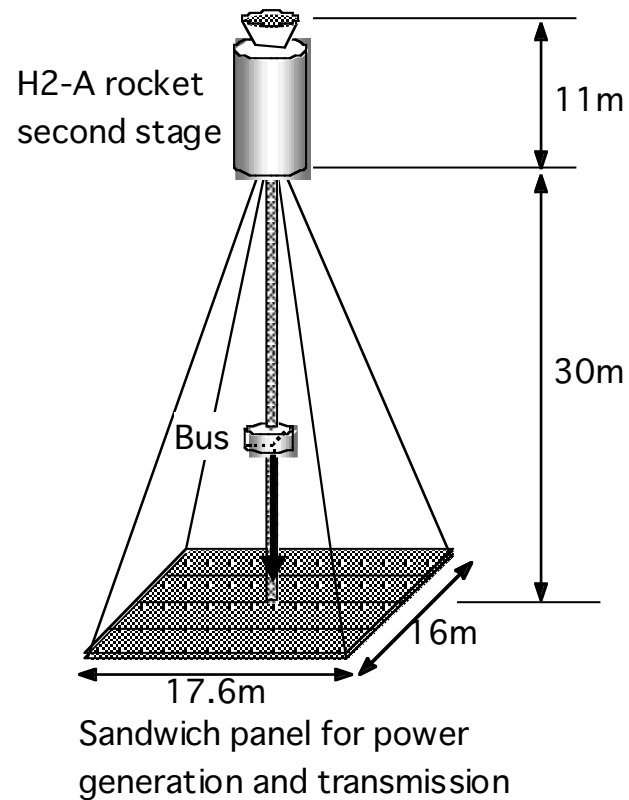
(2) Evaluation of over-all power efficiency as an energy system.

(3) Demonstration of electromagnetic compatibility with existing communication infrastructure.

(4) Study of operation procedure as an solar power satellite. 0



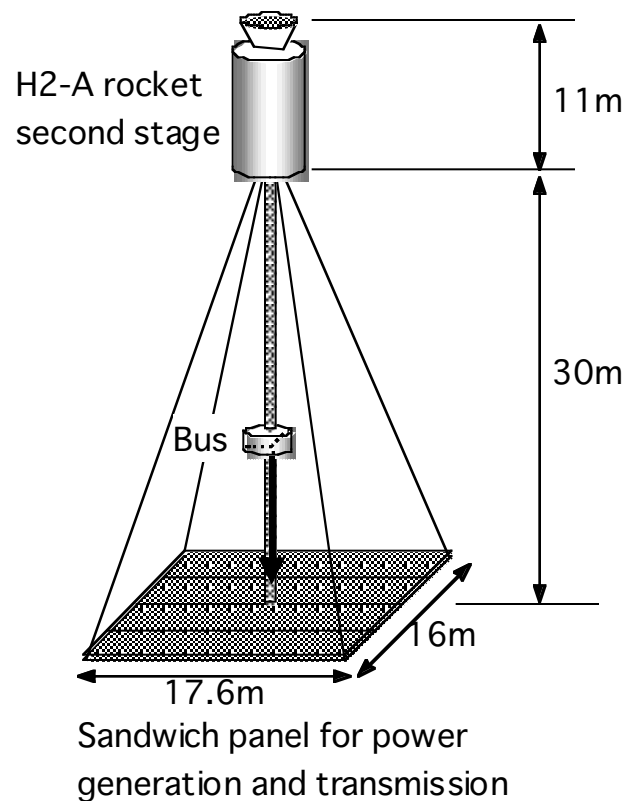
Configuration of Demonstration Experiment



- **Modular panel**(sandwich panel) combined with power generation and microwave transmission. 0
- **Small panels** folded in rocket and deployment in orbit to **a large single panel**. 0
- Attitude stabilization by gravity gradient force using a **tether system and truss system**. 0
- Thin film-type solar cells commercially available. 0
- Microwave transmission elements commercially available. 0
- **$f = 5.8\text{GHz}$** . 0
- Hybrid system combined with **magnetron and semiconductors**. 0
- **Wireless synchronization** of frequency and phase of oscillators for all panels. 0
- **Retrodirective beam control** using pilot signal from ground.

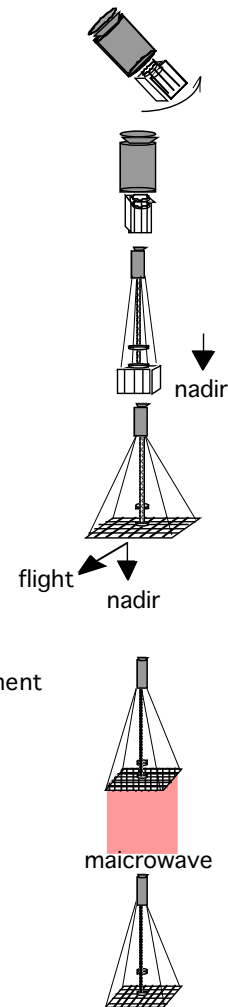
System of Demonstration Experiment 0

| | |
|----------------------|---|
| Size | 40.8m x 17.6m x 16.0m |
| Total Weight | 18,100 kg |
| Power Generation | 36 kW max |
| Power Transmission | 420 kW ~ 140 kW |
| Beam Control | retrodirective control |
| Microwave Frequency | 5.8 GHz |
| Operation | full power for 16 sec 10% power for 4 min |
| System Configuration | panel, truss, tether, weight mass |
| Panel Configuration | 80 foldable panels 400 power modules 250,000 antennas |
| Attitude Control | gravity gradient force |
| Altitude | 370 km |
| Rectenna type | parabola collector |
| Rectenna output | 30 kW ~ 10 kW (500 m diameter) |

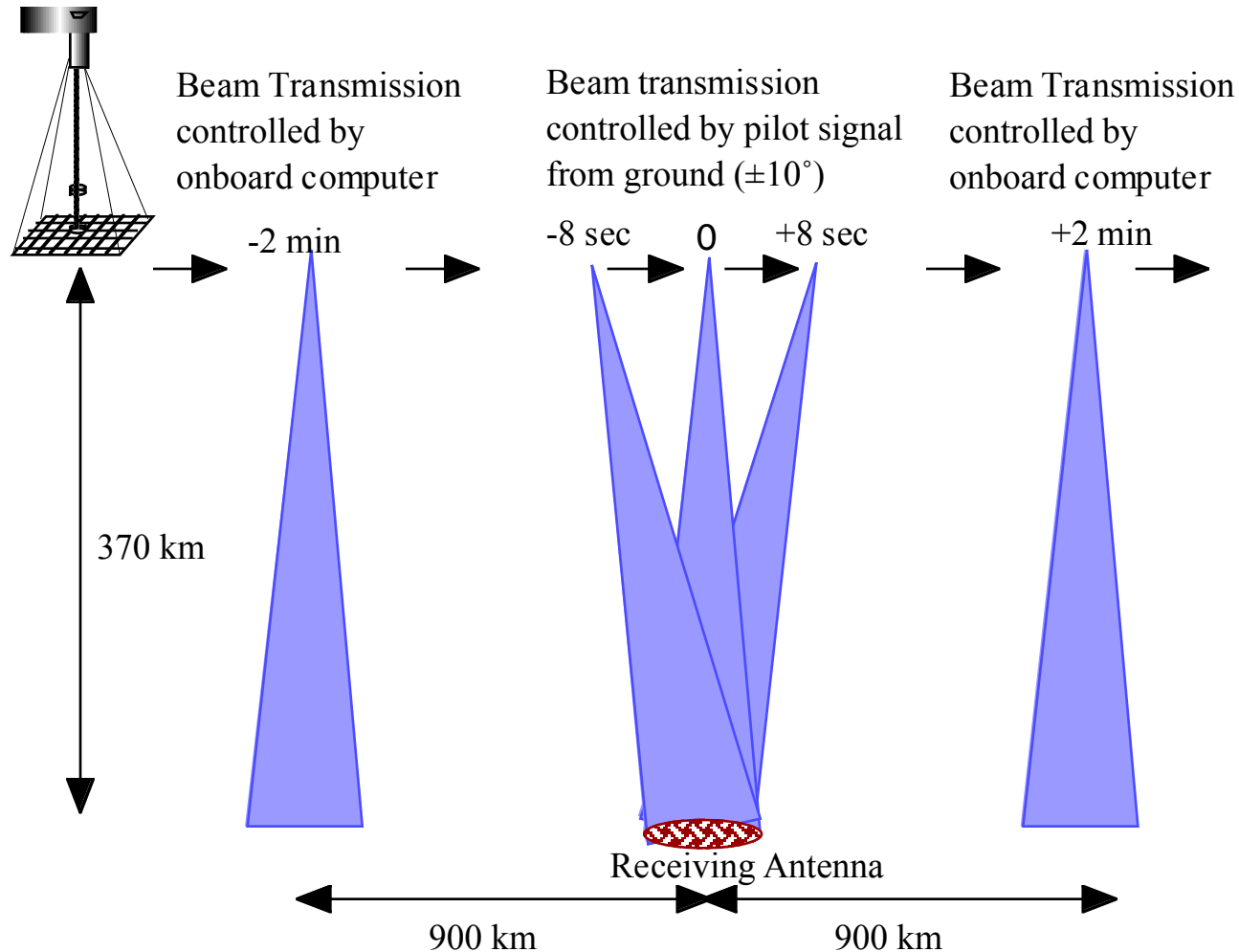


Mission Operation 0


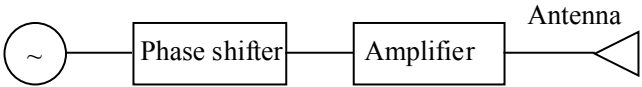
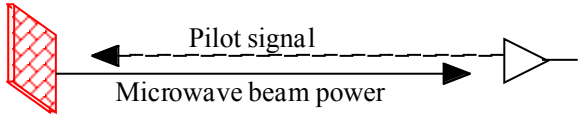
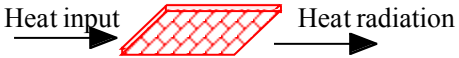
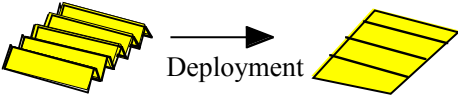
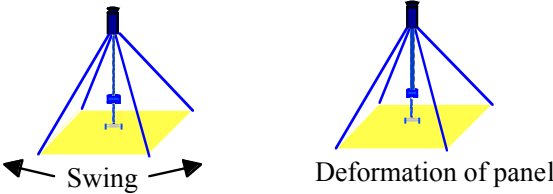
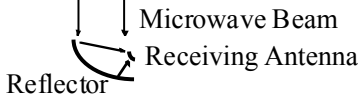
| | |
|-------------|---|
| M E T 0 | Launch Orbit (370km altitude) despin |
| M E T 1 | bus system checkout earth pointing truss deploiment |
| M E T 2 | panel deployment |
| M E T 3 ~ | orbit adjustment |
| M E T 3 0 | experiment checkout |
| M E T 3 1 ~ | power transmission experiment |
| | standby |



Experiment Sequence of Microwave Power Transmission 0



Critical Technologies for Demonstration Experiment 0

| Critical issues | Problem Area |
|---|--|
| Frequency and phase synchronization between sandwich panels |  <p>Wireless interface</p> |
| High-efficiency and low-loss microwave transmitter |  <p>Antenna</p> |
| Retrodirective beam control |  <p>Pilot signal</p> <p>Microwave beam power</p> |
| Thermal analysis |  <p>Heat input</p> <p>Heat radiation</p> |
| Two-dimensional panel deployment |  <p>Deployment</p> |
| Gravity stabilization by truss and tether hybrid system |  <p>Swing</p> <p>Deformation of panel</p> |
| Rectenna for low-power density microwave |  <p>Microwave Beam</p> <p>Receiving Antenna</p> <p>Reflector</p> |



Current Status of Research 0

| | Concept Development | Verification by simulation | Verification by experiment |
|---|---------------------|----------------------------|----------------------------|
| Frequency and phase synchronization between sandwich panels | | | |
| High-efficiency and low-loss microwave transmitter | | | |
| Retrodirective beam control | | | |
| Thermal analysis | | | |
| Two-dimensional panel deployment | | | |
| Gravity stabilization by truss and tether hybrid system | | | |
| Shape control of panel for microwave transmission | | | |
| Rectenna for low-power density microwave | | | |



Conclusion 0

- Demonstration experiment towards the practical SSPS has been studied.
- Sandwich panel module, wireless interface between modules, stabilization by gravity gradient force using tether and truss, are major characteristic features.
- Study indicates power transmission from space to ground more than 100kW or more is feasible.
- Further study is required for retro-directive beam control, two-dimensional panel deployment, and rectenna for low-power intensity microwave.